

### **C.) AMENDMENTS TO THE CLAIMS**

This listing of the claims will replace all prior versions and listings of the claims in the application.

1. (withdrawn) A ceramic matrix composite turbine engine component comprising:
  - a core insert section comprising a material selected from the group consisting of a silicon-silicon carbide composite and a monolithic ceramic; and
  - an outer shell section comprising a silicon carbide-silicon carbide composite material, wherein the at least one outer shell section and the at least one core insert section are bonded together using a silicon melt infiltration process.
2. (withdrawn) The turbine engine component of claim 1, wherein the component comprises a plurality of outer shell sections.
3. (withdrawn) The turbine engine component of claim 1, wherein the component comprises a plurality of core insert sections and a plurality of outer shell sections.
4. (withdrawn) The turbine engine component of claim 1, wherein the core insert section is a silicon-silicon carbide composite.
5. (withdrawn) The turbine engine component of claim 4, wherein the core insert section is manufactured using a Silcomp process.
6. (withdrawn) The turbine engine component of claim 1, wherein the core insert section is a monolithic ceramic material.
7. (withdrawn) The turbine engine component of 2, wherein the outer shell section is silicon carbide-silicon carbide composite material.
8. (withdrawn) The turbine engine component of claim 7, wherein the outer shell section is manufactured using a slurry cast melt infiltration process.
9. (withdrawn) The turbine engine component of claim 7, wherein the outer shell section is manufactured using a prepreg melt infiltration process.
10. (withdrawn) The turbine engine component of claim 1, wherein the component is a turbine vane.

11. (withdrawn) The turbine engine component of claim 1, wherein the component is a turbine nozzle.
12. (currently amended) A method of manufacturing a ceramic matrix composite turbine blade comprising the steps of:
  - providing a core insert section having a preselected geometry, the core insert section comprising a material selected from the group consisting of silicon carbide-silicon carbide composite preform having at least some porosity, silicon-silicon carbide composite, silicon-silicon carbide composite preform having at least some porosity, ~~silicon-silicon carbide composite~~, and a monolithic ceramic;
  - providing a plurality of plies of silicon carbide prepregged cloth;
  - laying up a preselected number of silicon carbide prepreg plies to form an outer shell section;
  - assembling the core insert section and the outer shell section into a turbine blade form, the turbine blade form comprising a dovetail section and an airfoil section, wherein the core insert section is positioned in the dovetail section of the turbine blade form;
  - autoclaving the turbine blade form;
  - filling remaining porosity in the turbine blade form with at least silicon using a silicon melt infiltration process, the filling also forming a bond between the core insert section and the outer shell preform.
13. (original) The method of claim 12, wherein the core insert section is a silicon-silicon carbide preform.
14. (original) The method of claim 13, wherein the silicon-silicon carbide preform includes carbon microspheres.
15. (original) The method of claim 12, wherein the core insert section is a silicon carbide-silicon carbide preform manufactured using a slurry cast process.
16. (original) The method of claim 12, wherein the core insert section is a silicon carbide-silicon carbide preform manufactured using a prepreg process.
17. (previously presented) A method of manufacturing a ceramic matrix composite turbine blade comprising the steps of:

providing a core insert section having a preselected geometry, the core insert section comprising a material selected from the group consisting of a silicon carbide-silicon carbide composite preform having at least some porosity, a silicon-silicon carbide composite, the silicon-silicon carbide composite preform having at least some porosity, and a monolithic ceramic;

providing an outer shell section preform, the outer shell preform having at least some porosity;

assembling the core insert section and the outer shell preform into a turbine blade form, the turbine blade form comprising a dovetail section and an airfoil section, wherein the core insert section is positioned in the dovetail section of the turbine blade form; and

filling remaining porosity in the turbine blade form with at least silicon using the silicon melt infiltration process, the filling also forming a bond between the core insert section and the outer shell preform.

18. (original) The method of claim 17, wherein the core insert section is a silicon-silicon carbide preform.
19. (original) The method of claim 18, wherein the silicon-silicon carbide preform includes carbon microspheres.
20. (original) The method of claim 19, wherein the core insert section is a silicon carbide-silicon carbide preform manufactured using a slurry cast process.